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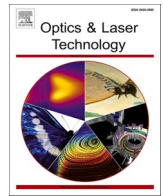
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Full length article

## Towards selective laser paint stripping using shock waves produced by laser-plasma interaction for aeronautical applications on AA 2024 based substrates

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### ABSTRACT

Laser stripping is a process which typically includes different forms of ablation phenomena. The presented work investigates a mechanical stripping process using high pressure laser-induced shock waves in a water confined regime. Power density is studied as a parameter for selective laser stripping on painted specimens and for adhesion relations with single layer epoxy targets. A flashlamp-pumped Nd:YAG laser with fixed spot size (4 mm) is shot on single layer epoxy and several layers of polymeric paint applied on a AA 2024-T3 (Aluminium) substrate. After laser treatment, samples are investigated with optical microscopy, profilometer and chemical analysis (FTIR & TGA). The results show that selective laser stripping is possible between different layers of external aircraft coatings and without any visual damage on the substrate material. In parallel to the experimental work, a numerical model has been developed to explain the background of the physical mechanisms and to qualitatively evaluate the detailed stress analysis and interfacial failure simulation for a single layer of epoxy on an aluminium substrate. The predicted failure patterns agree with the surfaces of the tested specimens observed by a microscope.

### 1. Introduction

Paint stripping is an essential process in the aeronautical industry since aircrafts need restoration, maintenance and routine inspection during their life time [1]. In the 1970s, the Air Force Logistics Command started to inspect alternative stripping methods to reduce the environmental drawbacks caused by the hazardous chemicals and waste disposed from the chemical stripping process. They found that plastic media blasting and laser stripping are the most applicable techniques mostly due to the environmental concerns. Eventhough the plastic media blasting found replaceable by the chemical stripping, there were still limitations on the use of it. While operating plastic media blasting,

the sensitive areas of the airplanes should be covered and the dust should be treated both during the blasting and after stripping operations. In addition, since plastic media blasting is not complete selective stripping process and in many cases damages the substrate surface (AA 2024-T3), laser stripping appeared as the most promising technique. Environmental friendliness, lower investment costs and the monitorable aspects of laser stripping process makes the application more favorable for industrial applications [2,3].

The external surface of aircrafts consists of a substrate with a surface treatment, a primer and a top coat. Substrates should be chosen from resistant materials like aluminium [4] with a surface treatment. The main function of the primer is to present a good level of adhesion

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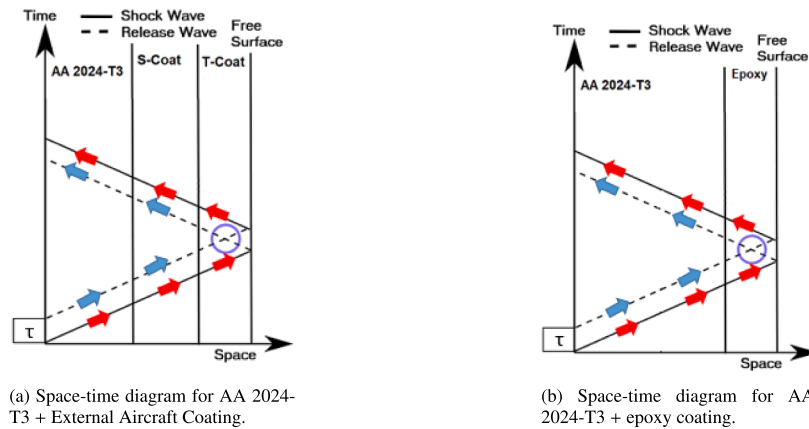


Fig. 1. Representation of shock wave propagation on space time diagrams for different layer configurations.

between layers and protecting the substrate from corrosion. In general, primers for aircrafts are made of epoxy components while top coats consist of polyurethane. The role of a top coat in aeronautical applications is to resist external environmental attacks (dust, sand, lightning strike) [5].

Laser paint stripping is common in the literature [6], however most research is focused on the paint ablation as a result of thermal effects such as Jasim *et al.* explained [7]. This paper presents a new stripping process which based on spallation phenomena produced by shock wave propagation via laser-plasma interaction from the back faces of specimens (aluminum side). This technique can be selective for thin coatings via (playing with the laser parameters) pure mechanical effects which is applied on a stack of AA 2024-T3 + External Aircraft Coatings (EAC) and AA 2024-T3 + Epoxy specimens.

The selective stripping capacity for polymeric painted layers as function of power density for AA 2024-T3 + EAC configuration is monitored. Moreover, laser stripping and adhesion relations on AA 2024-T3 + Epoxy are observed in order to obtain the process's capability from the back face applications.

This paper mainly consists of three parts. Part I describes the laser shock stripping process, part II presents methods and experimental procedure and part III demonstrates results and discusses them by focusing on the capability of the process to control the stripping depth as function of power density. Moreover, specific stripping phenomena is considered and compared with modeling via LS-DYNA software for single epoxy layer samples.

## 2. Process description

When a high intensity pulsed laser source (1 J, 10 ns) is focused on a target, high pressure plasma (GPa range) is generated and a shock wave is induced inside the specimen. For applications, confined regime with water is preferred for many years (see Fig. 2) [8]. Compared to the direct ablation regime (without any confinement), generated pressure on the target is two times longer and four times higher with the water confinement [9,10].

To present the physical principle, Fig. 1 shows a space-time diagram of stresses during the propagation of a representative shock wave produced by laser-plasma interaction with an acoustic approximation (pressure duration of " $\tau$ ") [11]. In Fig. 1a, S-coat (sub-coat) includes structural and exterior primers. In addition, T-coat (top-coat) consists of base and clear coat. Fig. 1b describes the case of an AA 2024-T3 with single epoxy layer. The straight lines represent the shock wave and the dotted lines correspond to the release wave. The created shock wave travels up to the back free surface of the specimens as shown with red arrows and then becomes a release wave after being reflected from the free surface. Blue arrays represent the initial release wave and the

reflected shock wave (reflection of the release). When the reflected release wave and the initial release wave meet, tensile stress creation occurs as depicted by a circle in Fig. 1 [12]. If it is high enough, damage of the material and/or interface debonding could occur which based on the current spallation phenomena [11]. By changing the laser parameters (pulse duration, number of applied pulses and different configurations of them), the tensile stress location can be modified at the desired depths.

These 1D diagrams broadly represent the basic stripping process using shock produced by laser-material interaction. However more complex 3D simulations are required to describe the physical mechanism as it will be seen later.

## 3. Materials and methods

### 3.1. General view

The applied procedure can be categorized as:

- Initial State Properties (Thickness measurements & Chemical Analysis)
- Laser Shock Tests (Stripping Threshold Evaluation)
- Post Mortem Analysis for Material States (Optical Microscopy, Profilometer & Chemical Analysis)
- Numerical Work (Observe Local Stress Levels)

In order to determine each layer's thickness (direct stack sizing) before any laser application, sample characterization is performed. To proceed the sample characterization, two methods are followed. One is performing the cut-off of samples to observe layer transitions and corresponding layer thicknesses. The second one is physico-chemical analysis (FTIR and TGA) which are used both before and after laser impacts to quantify the amount of unstripped layers. Since organic layers are thermally unstable, especially TGA appears as a pioneering method. Also, to determine the stripped layer thickness, a profilometer is utilized. The effect of the power density is measured on a single layer epoxy and several layers of polymeric paint on AA 2024-T3. Moreover, in order to observe the specific stress distribution along the focal spot for AA 2024-T3 + Epoxy samples, numerical work is done.

### 3.2. Material properties under shock using back face velocity analysis

Shocks produced via laser, generate high pressure (GPa) and high stress rate ( $10^6 s^{-1}$ ) within the material. To validate material modeling in these extreme conditions, mechanical testing is done in-situ. Well controlled laser loading (see part 3.3) is applied on the front surface of the target and material response is extracted from time resolved rear free

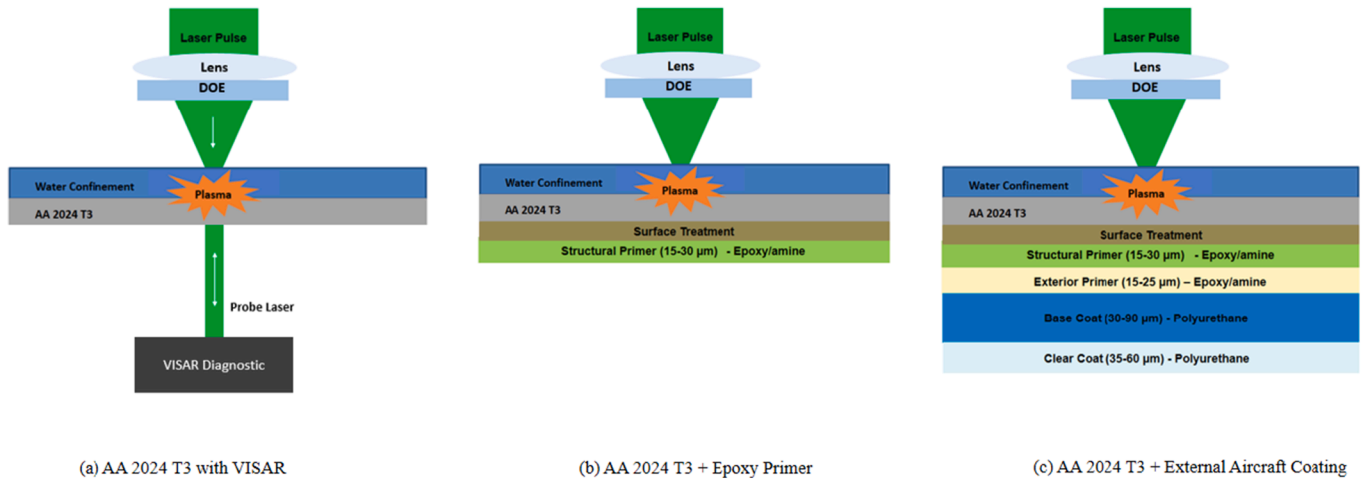


Fig. 2. Specimens Layer Configurations.



Fig. 3. Comparison of spatial shapes for 4 mm spot size, without & with DOE.

surface velocity. The material response measured by well known VISAR (Velocity Interferometer for Any System) diagnostic which is based on the Michelson Interferometer (see Fig. 2) [13 14]. It measures the Doppler Shift of mono mode probe laser (532 nm) that is reflected on the back face of the accelerated target via shock wave [15]. Hence, this accurate time-resolved measurement of the rear-free surface velocity gives an information about the propagation of shock waves and release waves from the top surface (plasma generation) to the rear surface (VISAR measurement).

In this paper, VISAR technique has been used to validate the Johnson Cook model implemented for AA 2024-T3 alloys (see Fig. 11) which has also done previously by Peyre *et al.* [16]. Calculations also give a realistic stress fields at the interface of the coatings, based on shock wave travel within the alloys (see part 5.1.3). Epoxy could not be tested by this technique since it is not reflective at the laser probe wavelength.

### 3.3. Laser set-up

Experiments were conducted at Hephaistos facility, PIMM lab, Paris, France. The laser used for the experiments is a Gaia HP laser from THALES company (France). It is a flashlamp-pumped Nd:YAG with a Gaussian temporal profile. The laser can produce 14 Joules of energy with 7.2 ns of pulse duration (FWHM) and 2 Hz of repetition rate at wavelength of 532 nm.

The obtained focal spots both with and without DOE (Diffractive Optical Element) obtained via camera (Camera Basler acA2040-25gm/gc, Monochrome, CMOS 1" with a Pixel Size of 5,5 μm × 5,5 μm) are represented in Fig. 3.

As one can see in Fig. 3a, the spatial shape is not homogeneous around the focal spot which is due to diffraction effect during the beam transportation and amplification. Hence there is a strong connection between the pressure and intensity profile, in order to apply an equal pressure through the whole focal spot on specimens, DOE is used (see

Table 1

Categorization of each sample with the thickness of each layer, respectively. CAA refers to Chromic Acid Anodization & CE is for Chemical Etching.

Base Material	AA 2024-T3 (970μ m) -A	AA 2024-T3 CAA (800μ m) -B	AA 2024-T3 CE (970μ m) -C	AA 2024-T3 CAA (800μ m) -D
Structural primer	-	Epoxy 37035A (28.5μ m)	Epoxy CA 7049 (28.1μ m)	Epoxy 37035A (28.2μ m)
Exterior Primer	-	-	-	Aerodur HS 2121 (27.4μ m)
Base Coat	-	-	-	Aerobase (44μ m)
Clear Coat	-	-	-	Clearcoat UVR (14-60μ m)

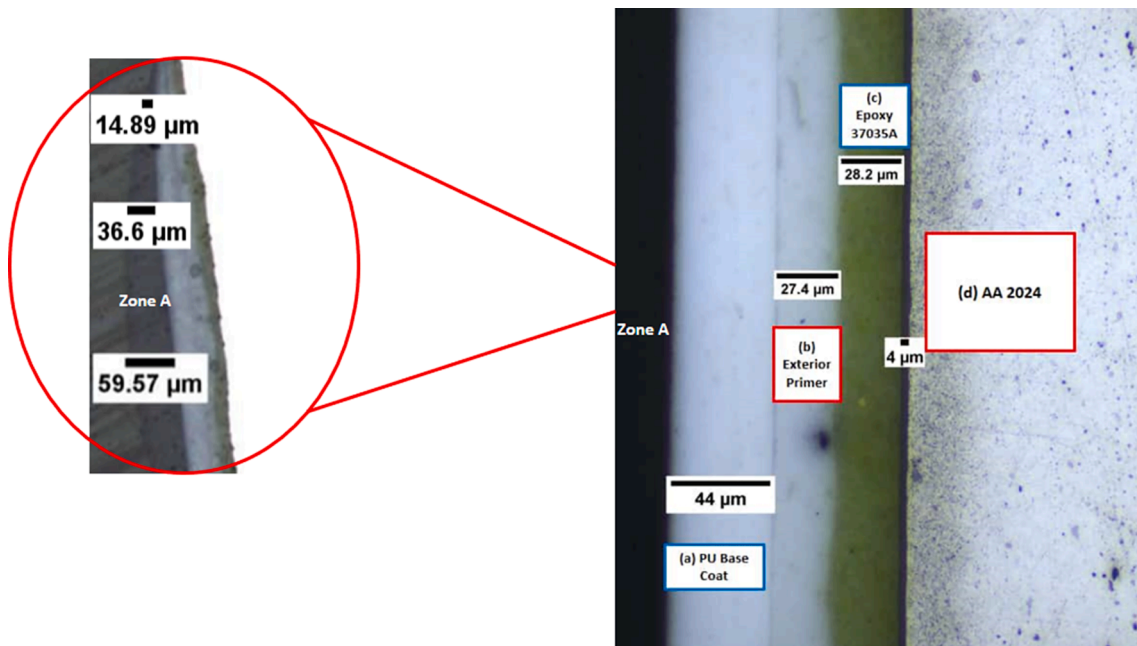
Fig. 3b) [12]. The 4 mm focal spot was obtained by a focal length of 198 mm lens.

For the performed experiments, mono impact (single beam) configuration is used. In addition, the energy output from the laser can be modified/controlled with the help of lambda quarter plate and polarizer rotated by motor. Energy measurement is done shot by shot and calibrated before each set of experiment using a calorimeter (QE50LP-H-MB-QED, Gentec, Québec, QC, Canada).

### 3.4. Samples

Three kind of samples are investigated which are described in Fig. 2, they were painted and prepared by manufacturers (RESCOLL, Akzo-Nobel & HAI). There are two different surface treatments for all sample types: Chromic acid anodization (CAA) and chemical etching (CE). The first sample type is AA 2024-T3 substrates which is used for VISAR experiments to monitor the shock wave behavior within the specimens (labeled as Sample A). The second type is AA 2024-T3 substrates with single epoxy layers with different surface treatments. Sample AA 2024-T3 CAA + Epoxy 37035A is labeled as Sample B and AA 2024-T3 CE + Epoxy CA 7049 is labeled as Sample C. The third type is AA 2024-T3 substrates with chromic acid anodization surface treatment plus External Aircraft Coatings (Sample D) as explained in Table 1. In order to determine the thickness of each layer, specimens are cut by a metallographic cut-off machine then observed by an optical microscope (ZEISS Axio Imager 2). Fig. 4 shows typical view for Sample D.

The thickness of the AA 2024-T3 is 800 and 970μ m for all configurations. For the Sample B, the thickness of the epoxy layer is as 28.5μ m and substrate surface has a chromic acid anodization surface treatment.



**Fig. 4.** Figure on the right is the cross-section cut of AA 2024-T3 CAA + EAC under the optical microscope. a) Polyurethane base coat, b) Exterior Primer, c) Structural Primer, 4 μm of surface treatment & the clear coat is labeled as Zone A. The figure left is the zoom-out of the cross section to demonstrate the thickness variation for the clear coat layer.

For Sample C, the epoxy thickness is  $28.1 \pm 4 \mu\text{m}$  with a surface treatment of chemical etching (all the stack sizing and descriptions are gathered in Table 1). Structural primer has a thickness range of  $28.2 \pm 4 \mu\text{m}$  and exterior primer is measured as  $27.4 \pm 2.5 \mu\text{m}$  for the investigated Sample D. Moreover, the polyurethane made base coat has a thickness of  $44 \pm 4.5 \mu\text{m}$  for analyzed samples. In addition, the final layer (clear coat) has a thickness in between 14.9 and  $60 \mu\text{m}$  as labeled as Zone A (Fig. 4). Since the clear coat is a transparent layer, we can not identify it without zooming out from the cross-section cut.

### 3.5. Depth profile analysis

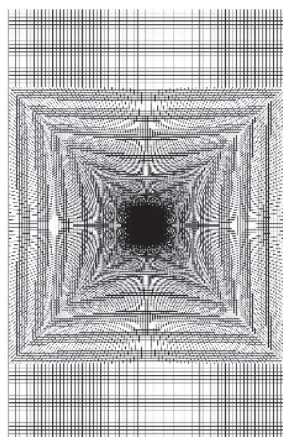
Depth of stripped coating for Sample B and Sample C have been investigated using optical profilometer Dektak 150. After applied laser impacts, stripped areas along the scanned focal spots are obtained. Since there are thickness variations especially for the clear coat as explained previously in Fig. 4, we can not expect them to have the same thickness for the whole sample which makes the measurement less accurate for the

samples that have more than one layer. As a result, depth profile measurements are not performed for Sample D after laser impacts.

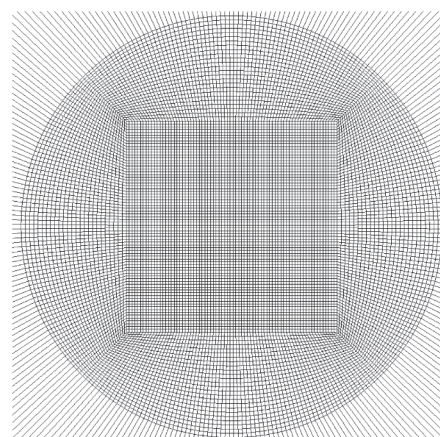
### 3.6. Physico-chemical characterization

FTIR (Fourier Transform InfraRed) spectra is collected in ATR (Attenuated Total Reflectance) mode using a Frontier 100 apparatus (Perkin Elmer) equipped with a diamond crystal and driven by the spectrum software. 16 scans are collected from  $650$  to  $4000 \text{ cm}^{-1}$  with a  $2 \text{ cm}^{-1}$  minimal resolution. FTIR analysis is performed to obtain signals for Sample B without any laser impact & Sample D after laser impact. The purpose of that is to obtain an information about the remained organic components on the aluminum substrate for Sample D after applied laser impacts in comparison with Sample B.

Thermogravimetric Analysis (TGA) is a method for monitoring the alteration of the weight against the temperature and time by certain atmospheres such as nitrogen and oxygen [17,18,19]. Moreover, the obtained weight loss is linked to different phenomenas. In the case of



(a) A typical FE mesh of the plate.



(b) A typical FE mesh of the load area (center of the plate).

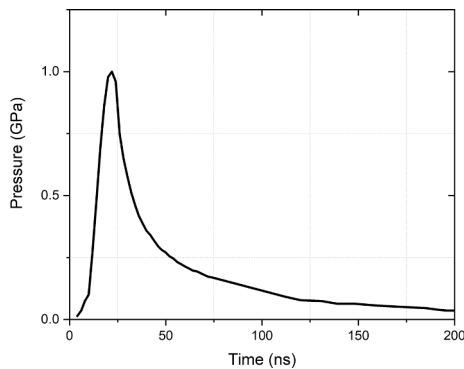
**Fig. 5.** 3D FE mesh model with a mesh size of  $5 \mu\text{m}$  through the thickness and  $45 \pm 15 \mu\text{mm}$  at the loaded area.

**Table 2**  
AA2024-T3 mechanical properties for Simplified Johnson Cook Model [28].

Symbol	Parameter	Value	Unit (SI)
$\rho$	Density	2700	kg/m <sup>3</sup>
E	Young's Modulus	73084	MPa
$\nu$	Poisson's ratio	0.33	
A	Strain Yield limit	352	MPa
B	Strain hardening modulus	440	MPa
n	Strain hardening exponent	0.42	
C	Strain rate coefficient	0.0083	

**Table 3**  
Elastic Plastic Hydrodynamic - Grüneisen for Epoxy Primer CA7049 [26].

Symbol	Parameter	Value	Unit (SI)
$\rho$	Density	1700	kg/m <sup>3</sup>
G	Shear Modulus	1600	MPa
$c_0$	Speed of the wave	2000	m/s
s	Linear Hugoniot slope coefficient	1.493	
$\gamma_0$	Material constant	1.13	



**Fig. 6.** The Pressure Profile as a function of time [29]

epoxies, only volatilization of degraded polymer is expected [20,21,22,23]. TGA (Thermogravimetric Analysis) was performed using a Q500 apparatus (TA Instruments) driven by TAQ series explorer. Tested samples were placed in platinum pans and were subjected to heating ramps from room temperature to 900 °C at a 10 °C min<sup>-1</sup> rate under 50 ml min<sup>-1</sup> nitrogen gas flow. Results were interpreted using TA Universal Analysis software.

#### 4. Numerical model

The 3D model has been developed using the explicit FE (Finite Element) software LS-DYNA. The dimensions of the aluminum/epoxy plate modeled are 80 mm × 125 mm × 0.97 mm and the thickness of the epoxy primer is 28 μm. The aluminum plate and the epoxy primer is modelled using 3D solid elements with one integration point (ELFORM = 1). The mesh size through the thickness of the material is 5 μm. Fig. 5a shows a typical FE mesh of the plate. As shown in Fig. 5b, for the load area, a denser mesh is used.

Failure of the materials' interface (stripping) is simulated using the CZM (Cohesive Zone Modeling) method. The interface is modeled using 4-point cohesive elements (ELFORM = 19) with zero thickness [24]. Damage initiation and propagation at the interface are simulated using a bi-linear mixed-mode I + II traction-separation law. This law follows an elastic behavior until a specific stress and after that point there is a degradation of the properties until final failure. The material model through which the CZM method is implemented in LS-Dyna the \*MAT\_138\_COHESIVE\_MIXED\_MODE. This material model needs

**Table 4**  
Pressure values for the corresponding laser intensities.

Power Density (GW/cm <sup>2</sup> )	Pressure (GPa)
1.75	2.72
2.28	3.00
8.23	7.12

separate parameters which are referred as fracture toughness properties of the interface. Due to lack of the required fracture toughness parameters of the aluminum/primer interface, the properties, which are used as input to the CZM method, belong to an adhesive with similar parameters to the epoxy primer.

To simulate the high strain rate dependent behavior of the aluminum and the hydrodynamic behavior of the epoxy, the simplified Johnson Cook plasticity model and the Grüneisen equation of state are used respectively. Implementation in LS-Dyna is done via \*MAT\_10\_ELASTIC\_PLASTIC\_HYDRO & \*MAT098\_SIMPLIFIED\_JOHNSON\_COOK material models respectively [25]. The elastic and hydrodynamic material properties [26,24,27,28] which are used as input to the material models, are presented in Table 2 and Table 3.

The applied boundary conditions restrict the displacement along z-axis (through the thickness) at the bottom right and left edge nodes. The pressure profile (see Fig. 6), is applied at the center of the front face of aluminum, with diameter of 4 mm. It is made to have a correlation between the experimental and the numerical results [29].

The peak value of each pressure profile that corresponds to a specific laser power density value, is calculated through experimental data. The applied pressure values for the power densities are given in Table 4.

## 5. Results and discussion

### 5.1. Stripping threshold

In order to observe the effect of an applied power density on laser stripping, laser impacts were applied from the back face of samples (aluminum side). Fig. 7 gathers direct observation analysis for each sample.

Results can be classified as : Power densities in which layers are removed/stripped (blue), debonded (green) and at initial state (purple).

From Fig. 7, for both painted samples and single layer epoxy specimens, stripping thresholds can be extracted as the smallest power density from which the first flight off of the coating is detected. While calculating the power densities, the laser's pulse duration, laser's energy and focal spot were the main parameters. As explained in part 3.3, the energy calibrated shot by shot and the laser pulse duration was constant as 7.2 ns. Hence, the main parameter creates a deviation for the power density values was the spot size. Spot size observations made with the camera (see part 3.3) and then measured with ImageJ software. After, the standard deviation is calculated for spot sizes and consequently for power density values. According to calculations there is ± 13% of relative standard deviation for mean power density values. For the presented work, used power densities are calculated for 4 mm spot size. Threshold values obtained as 2.77 GW/cm<sup>2</sup> for Sample D, 0.68 GW/cm<sup>2</sup> for Sample B and 1.75 GW/cm<sup>2</sup> for Sample C for 4 mm spot size. Mean power densities for 2.77 GW/cm<sup>2</sup> calculated as 3.04 GW/cm<sup>2</sup> (± 13%), for 0.68 GW/cm<sup>2</sup> as 0.75 GW/cm<sup>2</sup> (± 13%) and for 1.75 GW/cm<sup>2</sup> as 1.93 (± 13%) GW/cm<sup>2</sup> with the focal spot deviation.

#### 5.1.1. Visual analysis

Above the stripping threshold, the power density has been increased gradually at different locations on the target at 3.2 GW/cm<sup>2</sup>, 5.8 GW/cm<sup>2</sup> and 8.8 GW/cm<sup>2</sup> through 4 mm focal spot size. Then, samples have been investigated by optical microscope to observe the stripped surface status and layer characteristics, qualitatively. Fig. 8 presents the optical

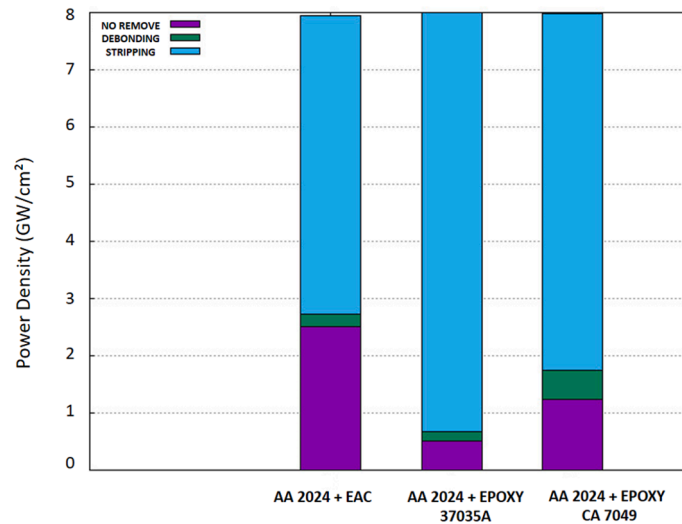


Fig. 7. Stripping Threshold as a function of power density. No stripping (purple), start of debonding (green) and stripping (blue).

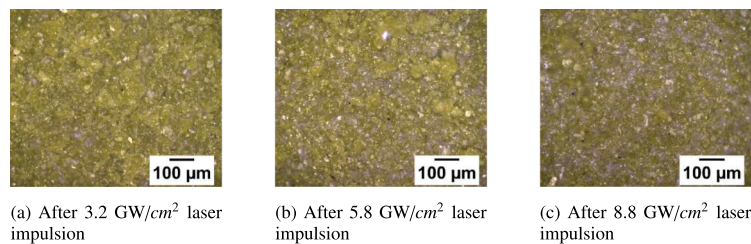


Fig. 8. Close-Up Optical Microscope Analysis of surfaces after laser impulsion for Sample D, respective power density values.

microscope analysis of Sample D specimen's surfaces after laser impacts where green layer is the structural epoxy primer (Epoxy 37035A layer).

As the surface appears visually more or less green (the structural primer color), these results show that, both at low and high applied power densities, structural primer stays present on the substrate. Therefore, the exterior primer coating + subcoating assembly are stripped for the Sample D type. However, with the increase of the power density, the structural primer presence is reduced on the bare surface. Eventhough, optical microscope analysis gives us a qualitative information about the layer status after laser impact, since there are thickness variations mostly due to clear coat thickness as explained previously, we can not expect them to have homogenously distributed thickness all around the sample. That's why instead of profilometer analysis, physico-chemical characterization has done for this sample type to obtain more quantitative data.

For Sample B and Sample C, partial stripping (damage ring) and complete stripping phenomenas were studied. With the increase of the power density, laser stripping status has been observed on specimens. Regardless of the type of an epoxy, surface treatment and base material's thickness, we obtained a common process window for the applied power density values. For Sample B, complete stripping was observed at 1.75  $\text{GW}/\text{cm}^2$  whereas for Sample C, partial stripping (damage ring) occurred at the same applied power density. Also, one can see that for Sample C at 2.28  $\text{GW}/\text{cm}^2$ , damage ring formation is occurred and for Sample B at 2.28  $\text{GW}/\text{cm}^2$  also damage ring formation observed in a different size. For both sample types, at 8.23  $\text{GW}/\text{cm}^2$ , the complete stripping occurred. In order to deeper investigate this phenomena, additional numerical work was followed for the cases of damage ring and complete stripping.

Fig. 9 shows the profilometer results for the stripped epoxy depth along the focal spot with corresponding microscope images for spots. Comparing the obtained depth vs scanning distance curves, we can have an idea about the surface health after the laser stripping process (since

the thickness of an epoxy layer is known before any laser impact). In addition, for partial stripped impacts, almost perfect top-hat shape was observed in the middle of the spots. As one can see from Fig. 9, in this power density range (for 1.75 & 2.28  $\text{GW}/\text{cm}^2$ ), the stripped/affected diameter becomes smaller with the increase of the power density. Whereas its 4080  $\mu\text{m}$  for Sample B at 1.75  $\text{GW}/\text{cm}^2$ , then it decreases to 3452  $\mu\text{m}$  at 2.28  $\text{GW}/\text{cm}^2$ . And for Sample C, the stripped/affected diameter is 3632  $\mu\text{m}$  at 1.75  $\text{GW}/\text{cm}^2$  and it then also decreases to 3588  $\mu\text{m}$ .

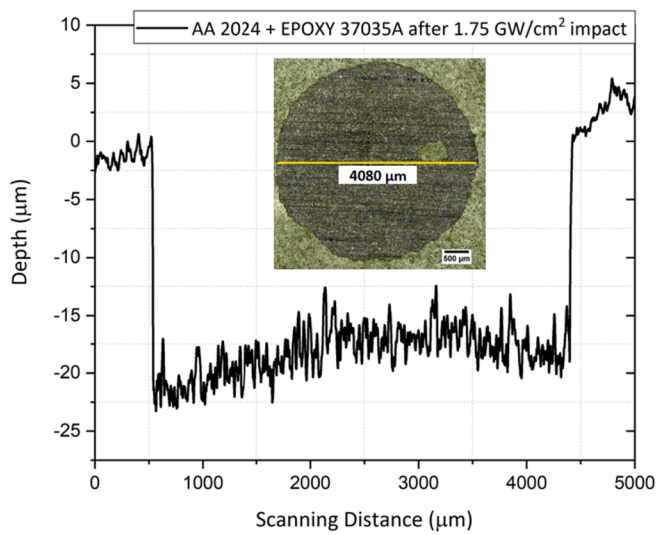
Since the epoxy thickness was 28.1–28.5  $\mu\text{m}$  as a result of the depth observations, the substrate was not damaged by the laser impacts except at the 2.28  $\text{GW}/\text{cm}^2$  for Sample B hence at this power density, the maximum depth reached more than 30  $\mu\text{m}$  (See Fig. 9b).

#### 5.1.2. Chemical analysis

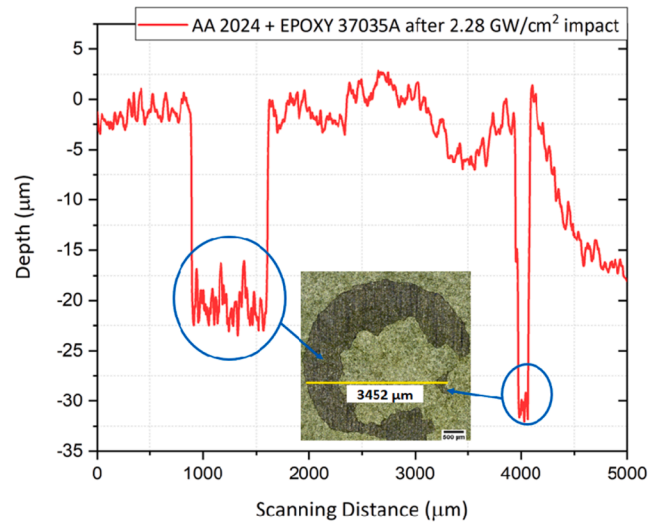
Thermogravimetric Analysis and Fourier Transform Infrared Spectroscopy analysis have been performed. The purpose of using TGA was to verify the remaining organic coatings after a laser impact in comparison with a single layer epoxy specimen and a sample which includes all external aircraft coatings. For Sample D and Sample B also the weight losses are correlated with the obtained stripping thresholds previously (see 5.1).

It is known that epoxies are thermally unstable and get decomposed at temperatures approximately above 300 °C. When decomposition is performed, there is no char residue [30,31] (contrarily to degradation under air). Also, polyurethane coatings would degrade at lower temperatures than epoxies [32,33,34]. Therefore, TGA has been used under pure nitrogen atmosphere as a simple method for quantifying the residual content of organic layers on the aluminum samples.

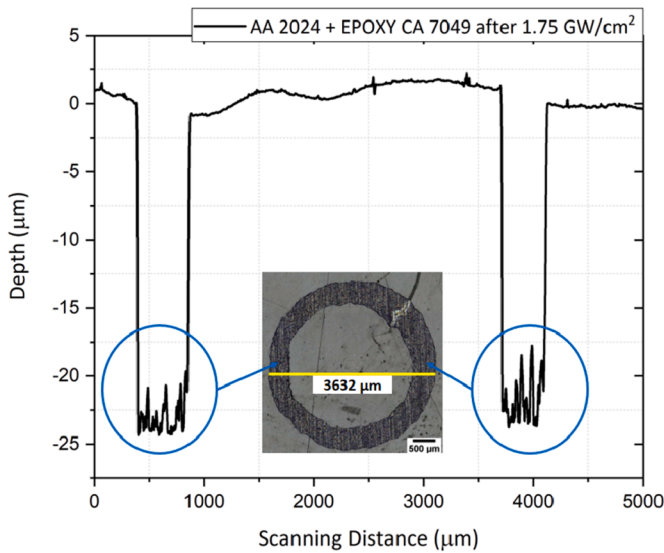
After obtained degradation curves which agreed well with the literature, the results linked with the laser stripping status and obtained maximum temperatures that corresponds to the temperature at which



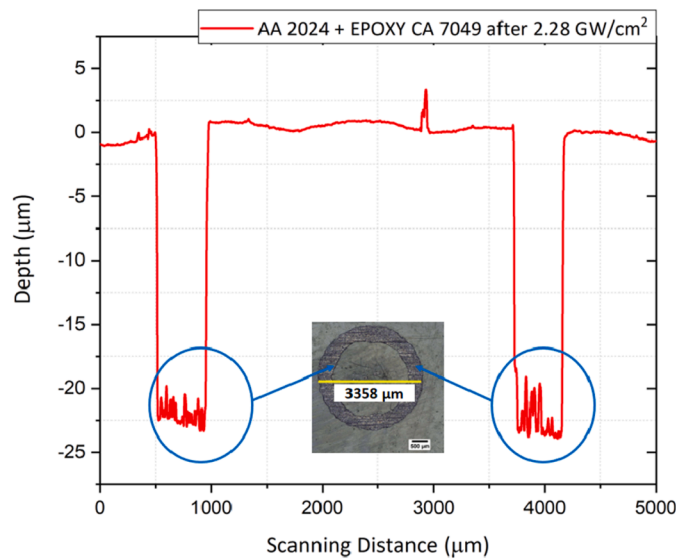
(a) AA 2024 + Epoxy 37035A after 1.75 GW/cm<sup>2</sup> impact.



(b) AA 2024 + Epoxy 37035A after 2.28 GW/cm<sup>2</sup> impact.



(c) AA 2024 + Epoxy CA 7049 after 1.75 GW/cm<sup>2</sup> impact.



(d) AA 2024 + Epoxy CA 7049 after 2.28 GW/cm<sup>2</sup> impact.

Fig. 9. Obtained depth profiles for different power density values and different type of epoxies.

**Table 5**  
Obtained weight losses and stripping thresholds.

Sample Name	Weight Loss (%)	Stripping Threshold (GW/cm <sup>2</sup> )	Maximum Temperature (°C)
Sample D	10.07	2.77	453.26
Sample D after 5.8 GW/cm <sup>2</sup> shot	3.42	–	472.88
Sample B	1.22	0.68	423

the greatest weight change occurs. The weight loss obtained by TGA was maximum (10.07%) with the highest stripping threshold for Sample D and minimum for Sample B (1.22%) with the lowest stripping threshold. For, Sample D after applied power density of 5.8 GW/cm<sup>2</sup> the weight loss value was in between those two samples (3.42%) (see Table 5).

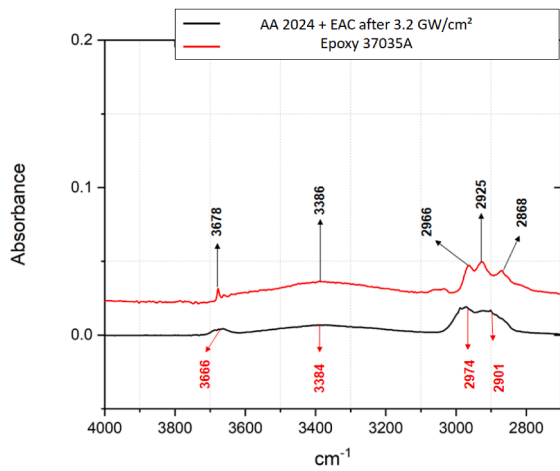
Eventhough, TGA is helpful for quantifying the remaining organic contents, in order to better understand the nature of the remaining structure, FTIR was utilized [35,36]. The purpose of using FTIR

diagnostic is to have a comparison between the single epoxy layer and the samples which already had a laser impact to see the remained organic coating type. First, the epoxy signature has been observed alone, then the comparison made with after applied laser impact surfaces.

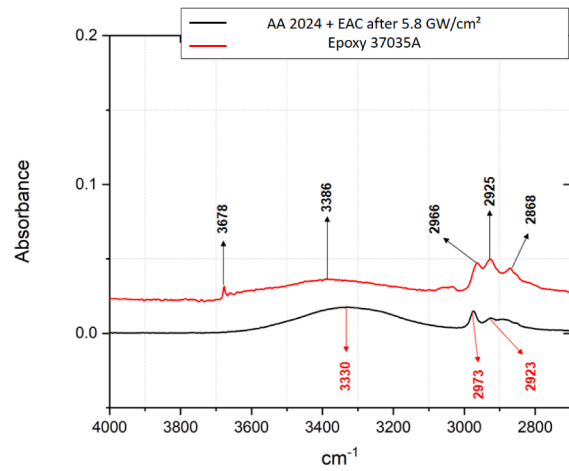
In order to better understand the nature of the remaining organic content structure, FTIR was utilized [35,36]. It is challenging to identify epoxy resins by FTIR since this family of thermoset contains a great amount of prepolymer-hardeners. However, in most cases, they can be tracked back by the presence of some specific groups such as phenyls (1610 cm<sup>-1</sup>) [37,38], secondary alcohols (2 hydroxy propyl ether) at (3400–3420 cm<sup>-1</sup>), isopropyl groups (2966 cm<sup>-1</sup>) [39], ethers (Ar-O-CH<sub>2</sub>) and alcohol (CH-OH) groups respectively at 1039 and 1109 cm<sup>-1</sup> [39].

Eventhough whole FTIR curves are obtained, only the range which is a typical absorbant for the epoxy is monitored. Since the Epoxy 37035A layer remained after 3.2 GW/cm<sup>2</sup> and 5.8 GW/cm<sup>2</sup> for Sample D according to the optical microscope observations, it was also verified by FTIR spectras (see Fig. 10). Although we can not conclude the result as





(a) FTIR Spectra for Epoxy and AA 2024-T3 CAA + EAC after 3.2 GW/cm<sup>2</sup> laser impact.



(b) FTIR Spectra for Epoxy and AA 2024-T3 CAA+EAC after 5.8 GW/cm<sup>2</sup> laser impact.

Fig. 10. Comparison of FTIR Spectras.

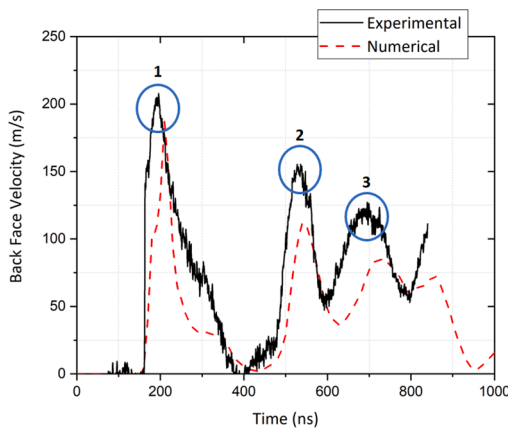


Fig. 11. Comparison of experimental and numerical velocities at 3 GW/cm<sup>2</sup>.

complete selective stripping, the comparison of these two different power densities underlined the more similar characteristic behavior for the impact of 3.2 GW/cm<sup>2</sup> with the single epoxy layer by taking into account the obtained peaks at (3666, 3384, 2974 and 2901 cm<sup>-1</sup>) see Fig. 10a. Hence corresponding peaks do not have a correlation for Sample D after 5.8 GW/cm<sup>2</sup> and Epoxy 37035A (see Fig. 10b).

### 5.1.3. Numerical results

5.1.3.1. *Model validation.* A first verification of the shock wave propagation simulation capability of the model was performed by comparing the numerical back-face velocities for a pure AA 2024-T3 specimen which has a thickness of 970 μm (Sample A) with an experimental curve constructed by VISAR measurements at 3 GW/cm<sup>2</sup>. The comparison is shown in Fig. 11. As shown, a good correlation has been achieved. The two graphs are close enough at the first peak and pull-back value but with a deviation at the second and third. The deviation was mainly attributed to the material model used and to the uncertainty in the correlation of the intensity with the exact applied pressure value.

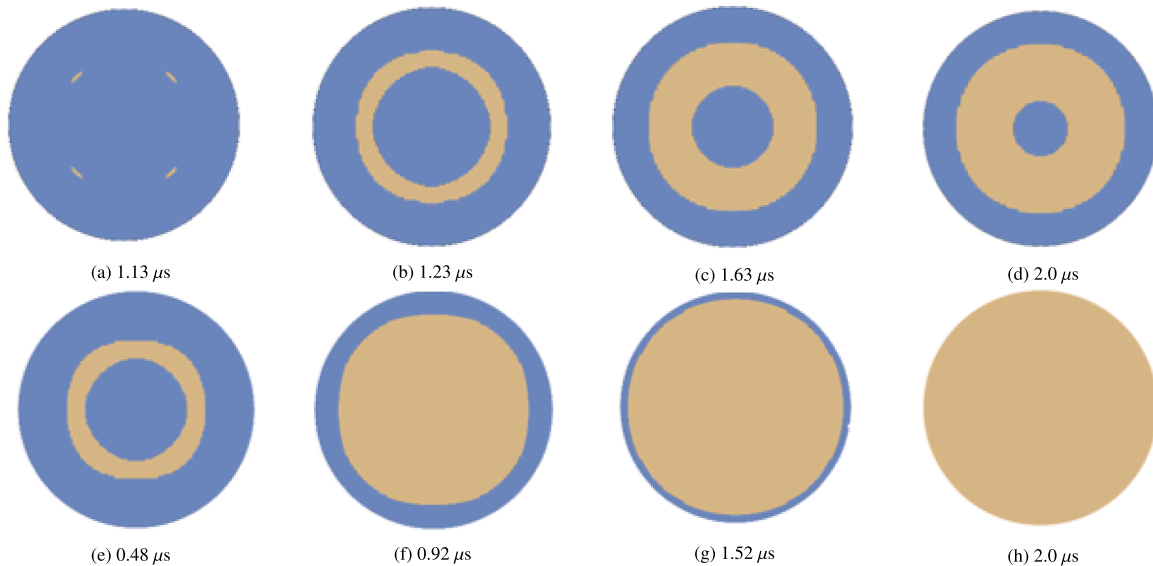


Fig. 12. Predicted initiation and propagation of failure of the cohesive zone for the cases of 1.75 GW/cm<sup>2</sup> (a,b,c,d) and 8.23 GW/cm<sup>2</sup> (e,f,g,h) power densities, respectively.

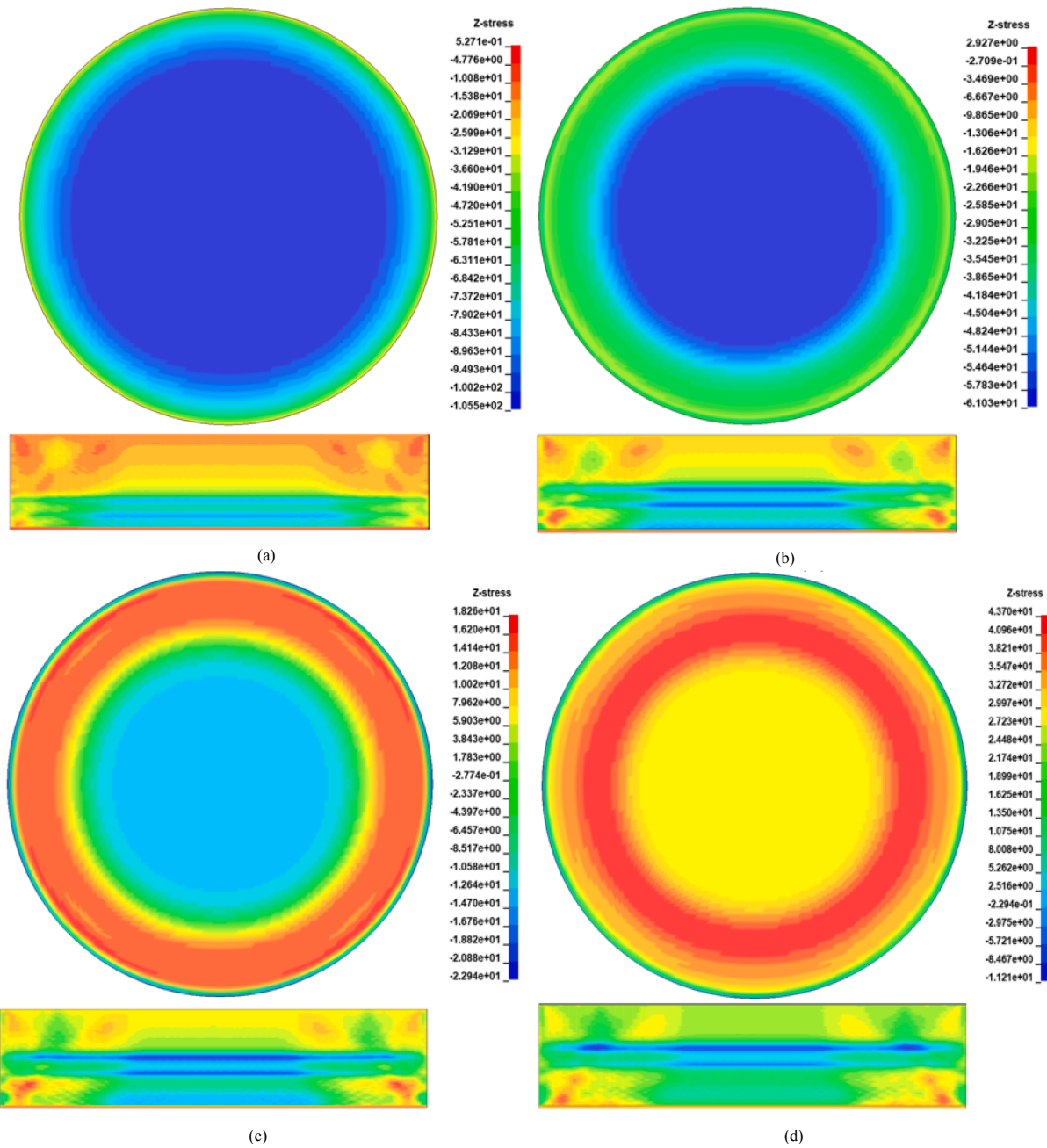


Fig. 13. Distribution of  $\sigma_z$  stresses on the epoxy at the interface (upper pictures) for various stages of shock wave propagation (lower pictures).

5.1.3.2. *Interfacial failure propagation.* Interfacial failure propagation was proceeded for the case of Sample B. The purpose of the findings was to couple them with the obtained experimental results. Fig. 12 shows the failure propagation as a function of time for 2 different power density values.

In Fig. 12, the predicted initiation and propagation of failure of the cohesive zone for the cases of  $I = 1.75 \text{ GW/cm}^2$  and  $I = 8.23 \text{ GW/cm}^2$  was depicted. Two different failure patterns were observed. For the lower intensity, a damage ring was formed (beige color represents the failure/stripping within the Sample B & the blue color represents the undamaged area) while for the higher intensity the entire circular area failed. The predicted final failure patterns for the two intensities agree with those observed experimentally, although a difference in the thickness of the damage ring was obtained for the lower intensity. This finding is a first verification of the model’s capability to simulate laser shock-induced stripping.

5.1.3.3. *Stress analysis.* In order to understand the failure process of the aluminum/epoxy interface, a detailed stress analysis was performed. The analysis was focused on the normal  $\sigma_z$  stresses, which are responsible for the stripping. Fig. 13 plots the contour of  $\sigma_z$  stresses developed on the surface of the epoxy at the interface at various stages of wave propagation (bottom pictures). The area where results are presented is the circular loading area. The compressive stresses can be distinguished by the blue color while tensile stresses by the red color. Fig. 13a shows when the compressive shock front reaches for the first time the epoxy, with a uniform stress field. With the propagation of the wave, this stress field tends to concentrate at the center thus forming an outer ring of weaker compressive stresses.

At this ring, the maximum tensile stresses were developed while at the center of the circle lower values of compressive stresses remained due to the development of a mach stem by the reflection of the shock front at the back free surface (see Fig. 14). Fig. 14 shows a schematic of

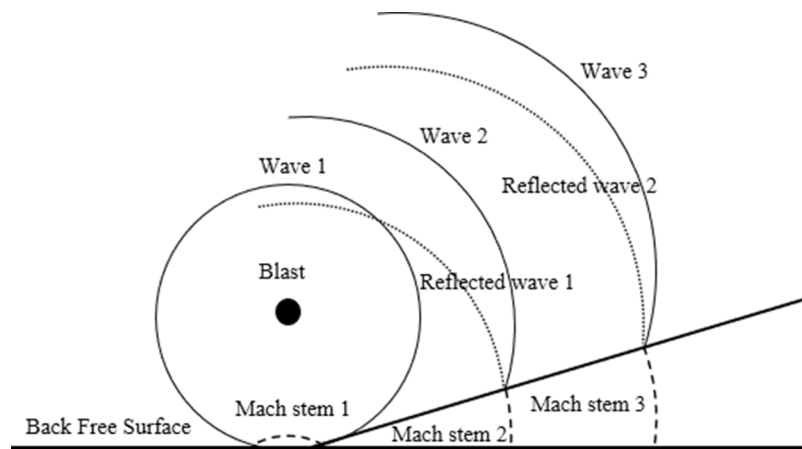


Fig. 14. Schematic of mach stem formation (corona).

the mach stem formation caused by the interaction of the reflection (dotted circle) of a shock wave (solid circle), at the back free surface, induced by a blast. When the mach stem gets weaker a ring of tensile stress, like before, tends to concentrate at the center of the specimen. This stress propagation pattern at the surface of epoxy continues for as long as shock propagates. Because of these stresses were developed at the surface of the epoxy and not in any solid volume there is no comparison with the yield stress limit or strength limit of the epoxy primer. In other words, no damage occurs due to the amplitude of the stresses because they don't correspond to stresses that can be compared with the strength limit or yield limit since they are at the surface and not in any volume, of the epoxy material.

## 6. Conclusions

This article investigated the mechanical stripping process of shock wave propagation created via laser-plasma interaction on various specimens of AA 2024-T3 with multiple painting stacks. The main parameter of interest is power density and how it affects the results in selective stripping for Sample D. Obtained results highlight that, with the modified power density values, laser stripping process is capable of removing both exterior primer coating + subcoating assembly. Optical microscope is used to examine the surface of specimens after laser impulse impacts. Fourier Transform Infrared Spectroscopy (FTIR) and Thermogravimetric Analysis (TGA) methods are used to analyse coating characteristics both before and after laser impulses. By coupling the TGA, FTIR, optical microscope and profilometer results, one might conclude that selective stripping can be achievable at certain applied power density values.

In addition, for Sample B and Sample C, adhesion relations are examined. The findings suggest that, regardless of the substrate thickness, the type of an epoxy and the surface treatments, partial stripping (damage ring) phenomena occurs. In order to understand the physical phenomena behind the partial stripping and to qualitatively evaluate the experimental findings, a numerical simulation of shock wave propagation into an aluminum/epoxy specimen is performed. The predicted interfacial failure patterns (annular and circular) agree well with the failure surfaces of the tested specimens. Moreover, the detailed stress analysis reveals that the distribution of stresses under the focal spot size is not homogeneous, which is explained by the mach stem formation phenomenon. The model, when fed with the actual material properties, is a promising contribution for future work that seeks to optimize the laser shock wave process and to conduct virtual testing.

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## Declaration of Competing Interest

The authors declared that there is no conflict of interest.

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